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DESIGN OF A SUPERVISED FLIGHT CONTROL SYSTEM FOR AIRCRAFT RELATIVE GUIDANCE

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Introduction

Considering the current and predicted levels of congestion of air traffic (see Figure 1), studies related to the delegation to the flight crew of some tasks currently performed by air traffic controllers are actively tackled today [1]. Among these studies, relative guidance between aircraft has appeared to be promising for the increase of air traffic capacity. The objective of this communication is to provide technical insight into the airborne devices and algorithms which may be used to automatically perform this new type of maneuver.

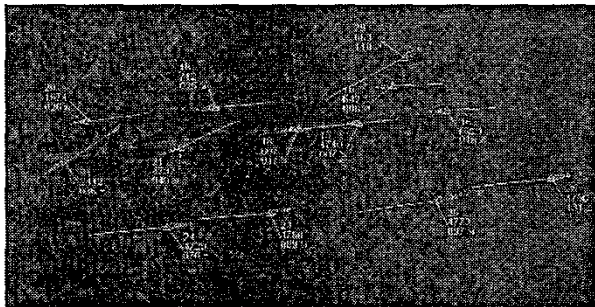


Figure 1. Radar Display at Paris-Charles de Gaulle

From an operational point of view, and assuming normal operations, the air traffic controller is relieved of providing instructions to the trailing aircraft for merging behind the leading aircraft and maintaining a given spacing once the flight crew has accepted a relative guidance clearance. Thus, the expected benefit of such new capabilities onboard aircraft is an increase of air traffic controller availability, which would result in increased air traffic capacity and/or safety. Enhancement of flight crew airborne traffic situational awareness is also expected with associated safety benefits. The feasibility of such a relative guidance device is based on the ability of

each aircraft to broadcast and receive suitable navigation data thanks to Automatic Dependent Surveillance-Broadcast (ADS-B) [2]. Among those navigation data, identification, position, speed and heading are of interest for the design of the relative guidance control law.

The literature dealing with aircraft relative guidance for Unmanned Air Vehicles (UAV) or military aircraft is quite large: in [3] and [4], the control device was designed on a linearized model, whereas in [5] the proposed feedback linearizing control law exhibits a singularity when the desired relative position is zero which may result in infinitely large inputs.

Nevertheless, research for civil aircraft in the relative guidance field is still in its initial stage. Indeed, performances of such aircraft are more constrained than those of military aircraft or UAV. In addition, safety and passenger comfort are crucial. In [6] station keeping is performed manually by the flight deck, whereas in [7] the authors consider a proportional, integral, and derivative (PID) control to control longitudinal station keeping. In addition, very few papers concentrate on the automatic control of the merging maneuver before maintaining the desired position behind the leading aircraft. Indeed, the merging maneuver exhibits large nonlinearities which cannot be handled by linearization. In [8] an approach based on flatness synthesis has been presented where the separation objective is expressed in terms of distance rather than delay.

In this paper, the automation of the relative guidance is performed thanks to a recursive non-linear control technique, namely backstepping [9]. This is a quite new design methodology for construction of both feedback control laws and associated Lyapunov functions in a systematic manner. In order to tackle safety and passenger

comfort aspects, which are crucial points for commercial aircraft, the controller is supervised: the purpose of this device is to handle on-line the gain of the controller, taking into account the current value of the state in order to satisfy some limitations in commercial aircraft maneuvering capabilities.

The paper is organized as follows: in the preliminary section, reference frame and aircraft model are introduced. This leads to a nonlinear state space representation. The subsequent section presents the design of the controller. Then, the supervision device is presented. Simulation results are provided in order to illustrate the approach. Finally, conclusions are raised.

Preliminaries

Separation Criteria

The separation to be applied between aircraft can be expressed as a distance, which is the current practice by air traffic controllers, or as a delay. This paper investigates a constant time separation. The interest of such a criteria is that limiting constraints such as runway occupancy, wake vortex decay and human reactions are naturally expressed in terms of time [10]. On the other hand, as current civil aviation regulations set distance separation standard between aircraft, the time delay separation objective must be chosen so that the minimum distance separation standard is not violated.

Aircraft Dynamics

As introduced in [3], it is assumed that the trailing aircraft is equipped with two autopilots which are able to operate in a decoupled fashion:

- Heading is assumed to be controlled through coordinated turns while roll and bank angle dynamics are neglected. Thus, the (small) bank angle change command, denoted φ_c , is related to the yaw rate $\dot{\psi}$ through the following relation, where g is the acceleration of gravity and V the actual airspeed :

$$\dot{\psi} = \frac{g}{V} \tan(\varphi_c) \approx \frac{g}{V} \varphi_c \quad (1)$$

- Airspeed is assumed to be controlled by auto throttle without affecting the aircraft's altitude through a first order linear model. Thus, the longitudinal acceleration \dot{V} is related to the commanded airspeed V_c , the current airspeed V and a time constant τ_V through the following relation :

$$\dot{V} = \frac{V_c - V}{\tau_V} \quad (2)$$

Kinematics of Relative Motion

This paper focuses on the simplified case where the aircraft relative motion is limited within the horizontal plane with no wind.

The nomenclature is indicated in the Figure 2.

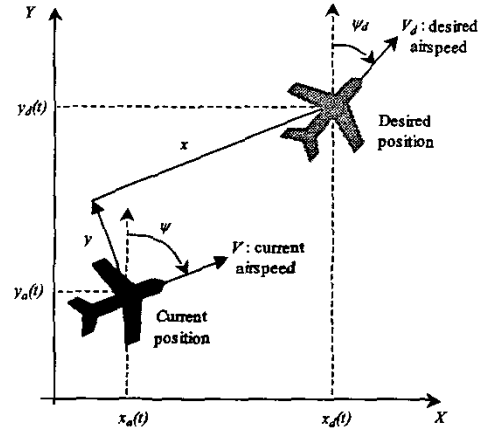


Figure 2. Nomenclature

- $x_a(t), y_a(t)$: the actual (or current) position of the trailing aircraft,
- $x_d(t), y_d(t)$: the desired position for the trailing aircraft. In the context of constant time separation, the desired position is the position of the leading aircraft ninety or so seconds before the current time,
- V, V_d : respectively, the current and the desired airspeed,
- ψ, ψ_d : respectively, the current and the desired heading,
- x, y : respectively, the along-track distance and the cross-track distance between the actual and the desired position, expressed in the reference frame affixed to the current position.

With the notation shown in the previous figure, the equations of relative motion between the actual and desired positions are found to be:

$$\begin{bmatrix} \dot{x} \\ \dot{y} \end{bmatrix} = \begin{bmatrix} 0 & \dot{\psi} \\ -\dot{\psi} & 0 \end{bmatrix} \begin{bmatrix} x \\ y \end{bmatrix} + \begin{bmatrix} -V + V_d \cos(\psi - \psi_d) \\ -V_d \sin(\psi - \psi_d) \end{bmatrix} \quad (3)$$

Controller Design

Backstepping is a quite new recursive design methodology for construction of both feedback control laws and associated Lyapunov functions. It was introduced by Krstic, Kanellakopoulos and Kokotovic [9]. Backstepping applies on cascaded nonlinear systems and, contrary to feedback linearization, it allows to retain stabilizing nonlinearities in the control design. This paper considers a vectorial backstepping technique which exploits a skew-symmetric property of a matrix which appears in the relative motion kinematics.

The state space representation of the system is obtained by gathering (1), (2) and (3):

$$\dot{\underline{x}}_1 = a(\underline{x}_2, \underline{u}) \underline{x}_1 + b(\underline{x}_2) \quad (4)$$

$$\dot{\underline{x}}_2 = c(\underline{x}_2) + d(\underline{x}_2) \underline{u} \quad (5)$$

Where:

$$\begin{cases} \underline{x}_1 = \begin{bmatrix} x & y \end{bmatrix}^T \\ \underline{x}_2 = \begin{bmatrix} \psi & V \end{bmatrix}^T \\ \underline{u} = \begin{bmatrix} \varphi_c & V_c \end{bmatrix}^T \end{cases} \quad (6)$$

And:

$$\begin{cases} a(\underline{x}_2, \underline{u}) = \begin{bmatrix} 0 & g\varphi_c/V \\ -g\varphi_c/V & 0 \end{bmatrix} \\ b(\underline{x}_2) = \begin{bmatrix} -V + V_d \cos(\psi - \psi_d) \\ -V_d \sin(\psi - \psi_d) \end{bmatrix} \\ c(\underline{x}_2) = \begin{bmatrix} 0 \\ -V/\tau_v \end{bmatrix} \\ d(\underline{x}_2) = \begin{bmatrix} g/V & 0 \\ 0 & 1/\tau_v \end{bmatrix} \end{cases} \quad (7)$$

The design objective is to render the equilibrium point ($\underline{x}_1 = \underline{0}$; $\underline{x}_2 = \underline{x}_{2d}$) globally asymptotically stable. Since the nonlinear systems (4) and (5) consist of two states \underline{x}_1 and \underline{x}_2 , and taking into account the fact that the matrix $a(\underline{x}_2, \underline{u})$ which appears in (4) is skew-symmetric, the vectorial backstepping technique can be applied by considering systems (4) and (5) as two cascaded systems:

- system (4) has $b(\underline{x}_2)$ as virtual input, and \underline{x}_1 as output,
- system (5) has \underline{u} as input, and \underline{x}_2 as output.

For system (4), the virtual control $b(\underline{x}_2)$ is chosen in order to stabilize it around $\underline{x}_1 = \underline{0}$; by denoting Λ_1 a positive definite matrix (tuning parameter) and by \underline{z} a stabilizing function which provides feedback to the system, the virtual control is chosen as follows:

$$b(\underline{x}_2) = \underline{z} - \Lambda_1 \underline{x}_1 \quad (8)$$

In a second step, let us consider the following candidate Lyapunov function, where k_1 is a positive number (tuning parameter):

$$V = \frac{k_1}{2} \underline{x}_1^T \underline{x}_1 + \frac{1}{2} \underline{z}^T \underline{z} \quad (9)$$

Taking into account (8) and the skew-symmetric property of the matrix $a(\underline{x}_2, \underline{u})$, the time derivative of (9) leads to :

$$\dot{V} = -k_1 \underline{x}_1^T \Lambda_1 \underline{x}_1 + \underline{z}^T (k_1 \underline{x}_1 + \Lambda_1 \dot{\underline{x}}_1 + \nabla b(\underline{x}_2) \dot{\underline{x}}_2) \quad (10)$$

Where $\nabla b(\underline{x}_2)$ is the Jacobian matrix defined by:

$$\nabla b(\underline{x}_2) \triangleq \frac{\partial b(\underline{x}_2)}{\partial \underline{x}_2^T} \quad (11)$$

Finally, the control vector \underline{u} is chosen by expanding $\dot{\underline{x}}_2$ in (10) thanks to (5) so that the time derivative of the candidate Lyapunov function is negative. By denoting Λ_2 a positive

definite matrix (tuning parameter), this leads to the following:

$$\underline{u} = (\nabla b(\underline{x}_2) d(\underline{x}_2))^{-1} (-k_1 \underline{x}_1 - \Lambda_1 \dot{\underline{x}}_1 - \nabla b(\underline{x}_2) c(\underline{x}_2) - \Lambda_2 \underline{z}) \quad (12)$$

To be more specific, let us select Λ_1 and Λ_2 as diagonal matrix:

$$\begin{cases} \Lambda_1 = \text{diag}(\lambda_x, \lambda_y) \\ \Lambda_2 = \text{diag}(\lambda_v, \lambda_\psi) \end{cases} \quad (13)$$

Thus, control law (12) expands as follows:

$$\begin{cases} \varphi_c = \frac{V}{g(V_d \cos(\psi - \psi_d) + \lambda_y x)} \left\{ (k_1 + \lambda_y \lambda_\psi) y - (\lambda_y + \lambda_\psi) V_d \sin(\psi - \psi_d) \right\} \\ V_c = V + \tau_v \left\{ (\lambda_x + \lambda_v)(V_d \cos(\psi - \psi_d) - V) + (k_1 + \lambda_x \lambda_v) x + \frac{g}{V} \varphi_c (\lambda_x y - V_d \sin(\psi - \psi_d)) \right\} \end{cases} \quad (14)$$

Supervision of the Controller

The purpose of the controller supervisor is to handle on-line the gain of the controller, taking into account the current value of the state in order to comply with some limitations in commercial aircraft maneuvering capabilities. This is illustrated in the Figure 3.

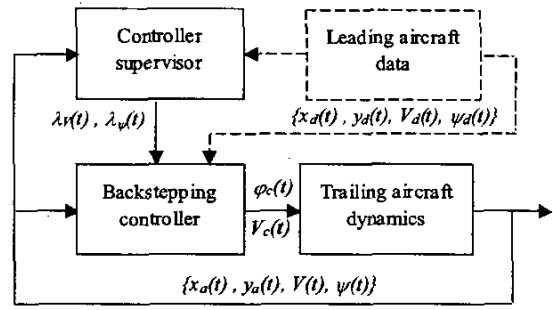


Figure 3. Supervised Controller

To be more specific, this paper focuses on the following constraints:

- to give favor to constant speed maneuver when the cross-track distance between the trailing and the leading aircraft is 'high'. The idea behind this constraint is to do 'one thing at a time' in order to enable the flight crew to understand the controller behavior. As a consequence, the merging maneuver should start at constant airspeed as far as the leading aircraft is not in the line of sight of the trailing aircraft. Then, when the trailing aircraft is established behind the leading aircraft, airspeed may vary,
- to provide accelerations which are acceptable from a passenger point of view,
- to maintain bank angle lower than the maximum bank angle of operation,
- to maintain the airspeed of the trailing aircraft between minimum and maximum speed of operation.

In order to satisfy the preceding constraints, the tuning parameters k_1 , Λ_1 and Λ_2 shall be set dynamically. First of all, it is worth noticing that k_1 and Λ_1 have been assumed to be constant during the design process, whereas $\Lambda_2 = \text{diag}(\lambda_v, \lambda_\psi)$ may depend on time. In addition, expression (14) shows that parameter λ_v just appears within the expression of the commanded speed V_c , whereas parameter λ_ψ just appears within the expression of the commanded bank angle φ_c . Thus, the idea is to set λ_v and λ_ψ so that it compensates exactly for k_1 and Λ_1 in order to reduce the number of parameters to be tuned. To achieve this goal, let us select $\lambda_v(t)$ and $\lambda_\psi(t)$ as follows:

$$\begin{cases}
 \lambda_v(t) = \lambda_{v_0}(t) - \frac{1}{V_d \cos(\psi - \psi_d) - V + \lambda_x x} \left(k_1 x + \lambda_x (V_d \cos(\psi - \psi_d) - V) + \frac{g}{V} \varphi_c (\lambda_x y - V_d \sin(\psi - \psi_d)) \right) \\
 \lambda_\psi(t) = \lambda_{\psi_0}(t) - \frac{1}{\lambda_y y - V_d \sin(\psi - \psi_d)} \left(k_1 y - \lambda_y V_d \sin(\psi - \psi_d) \right)
 \end{cases} \quad (15)$$

Introducing those values for $\lambda_v(t)$ and $\lambda_\psi(t)$ into (14) leads to the following expressions for the commanded bank angle and airspeed where parameter k_1 has disappeared:

$$\begin{cases}
 \varphi_c = \frac{V(\lambda_y y - V_d \sin(\psi - \psi_d))}{g(V_d \cos(\psi - \psi_d) + \lambda_y x)} \lambda_{\psi_0}(t) \\
 \frac{V_c - V}{\tau_v} = (V_d \cos(\psi - \psi_d) - V + \lambda_x x) \lambda_{v_0}(t)
 \end{cases} \quad (16)$$

To give preference to maneuver with no longitudinal acceleration when the cross-track distance $y(t)$ is 'high', $\lambda_{\psi_0}(t)$ is set as follows:

$$\lambda_{\psi_0}(t) = \lambda_{\psi_0}(0) \exp(-\alpha_0 |y(t)|) \quad (17)$$

The constraint dealing with bank angle will be treated through parameter λ_ψ . As a consequence, $\lambda_{\psi_0}(t)$ is set as a constant:

$$\lambda_{\psi_0}(t) = \lambda_{\psi_0}(0) \quad (18)$$

In the following, we denote :

- V_{min} the minimum speed of operation,
- V_{max} the maximum speed of operation,
- a_{max} the maximum acceleration acceptable from a passenger point of view,
- φ_{cmax} the maximum bank angle of operation.

In order to satisfy the acceleration and the bank angle constraints, it is clear from relations (16) that parameters λ_x and λ_y shall be chosen as a function of λ_{v_0} and λ_{ψ_0} as follows:

$$\begin{cases}
 0 < \lambda_y \leq \frac{1}{y_{max}} \left(g \frac{V_{dmin}}{V_{max}} \frac{\varphi_{cmax}}{\lambda_{\psi_0}(0)} - V_{dmax} \right) \\
 0 < \lambda_x \leq \frac{1}{x_{max}} \left(\frac{a_{max}}{\lambda_{v_0}(0)} - (V_{dmin} + V_{max}) \right)
 \end{cases} \quad (19)$$

Finally, the airspeed of the trailing aircraft is maintained between the minimum and maximum speed of operation by limiting the outputs of the controller.

Simulations

Scenario

In this section, a scenario is designed in order to evaluate the properties of the control laws previously designed. A special attention is given on the behavior of such a controller to the changes of heading and airspeed from the leading aircraft.

The leading aircraft starts at $x_0 = 0$ NM, $y_0 = 0$ NM, with initial conventional airspeed and heading of 200 kts and 90 degrees respectively. It is supposed to broadcast position, airspeed and heading every second. No wind is assumed.

The controlled bank angle of the leading aircraft is always zero, except between 220 sec and 310 sec where the leader changes its heading of about 155 degrees with a bank angle of 20 degrees.

The controlled conventional airspeed of the leading aircraft is first set at 220 kts for $t \leq 400$ sec, and then is set to 160 kts.

The trailing aircraft starts at $x_0 = -8$ NM, $y_0 = +4$ NM, with initial conventional airspeed and heading of 220 kts and 90 degrees respectively.

The simulation lasts 15 min (900 sec), and the requested time based separation for the trailing aircraft is constant and equal to 90 sec behind the leading aircraft.

During the maneuver, the outputs (i.e. the controlled bank angle and airspeed) of the relative guidance controller are limited to the following 'safe' values:

$$\begin{cases} -20 \text{ deg.} \leq \varphi_c \leq +20 \text{ deg.} \\ 140 \text{ kts} \leq V_c \leq 250 \text{ kts} \end{cases} \quad (20)$$

The time constants τ_v and τ_ϕ of the airspeed and heading autopilots are set to the following values:

$$\begin{cases} \tau_v = 40 \text{ sec} \\ \tau_\phi = 5 \text{ sec} \end{cases} \quad (21)$$

The values of the gains for the controller and the supervisor have been set as follows:

$$\begin{cases} \lambda_{v_0}(0) = 1 \text{ sec}^{-1} \\ \alpha_0 = 5 \text{ NM}^{-1} \\ \lambda_{\psi_0}(0) = 0.5 \text{ sec}^{-1} \\ \lambda_x = 0.01 \text{ sec}^{-1} \\ \lambda_y = 0.01 \text{ sec}^{-1} \end{cases} \quad (22)$$

Movement in the Horizontal Plane

The movements of the leading and trailing aircraft in the horizontal plane are shown in the following figure. As expected, the trailing aircraft move towards the leading aircraft (See Figure 4).

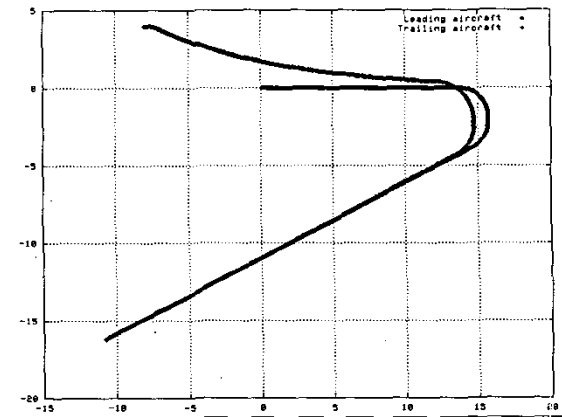


Figure 4. Movement of the Trailing and Leading Aircraft in the Horizontal Plane (Axes in NM)

Range

The evolution of the actual range between the leading and the trailing aircraft is shown in the following figure. Before leading aircraft speed change, the desired slant range is 5.5 NM, i.e. $220 \text{ kts} \times 90 \text{ sec}$. Due the heading change of the leading aircraft between 220 sec and 310 sec, the value of the slant range decreases to about 4.5 NM before starting to increase. Finally, when the leading aircraft is established on the final straight line at constant speed, the slant range reduces as intended to 4 NM, i.e. $160 \text{ kts} \times 90 \text{ sec}$ (See Figure 5).

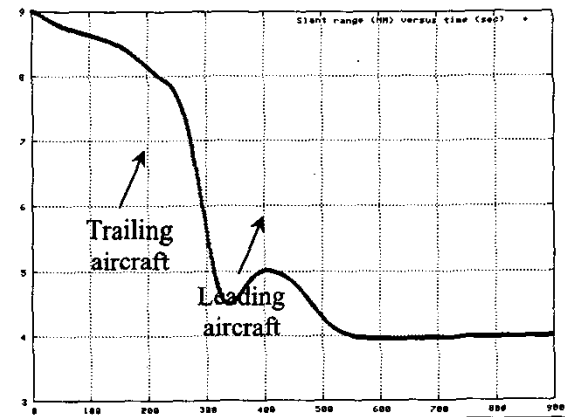


Figure 5. Slant Range (NM) between the Leading and the Trailing Aircraft versus Time (Sec)

Airspeed

The evolution of the actual airspeeds of the leading and the trailing aircraft is shown in the following figure. As designed, the airspeed is constant, 220 kts, when the cross-track distance between the trailing and the leading aircraft is 'high'. At about 2 NM of cross-track distance, the airspeed starts to increase up to the maximum operation airspeed, i.e. 250 kts. The change in heading for the leading aircraft induces the decrease to about 205 kts before a small increase. Finally, the decrease towards 160 kts of the airspeed of the leading aircraft induces the decrease of the airspeed of the trailing aircraft towards the same value (see Figure 6).

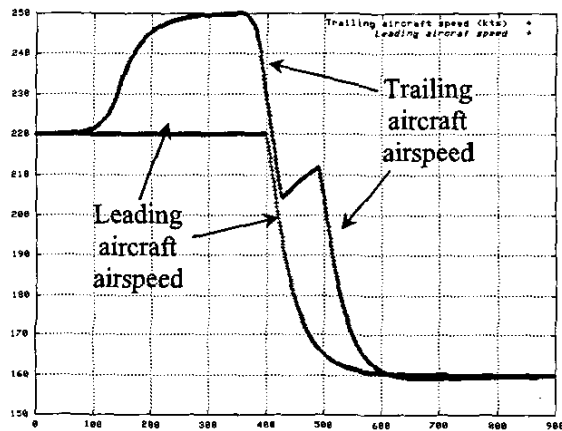


Figure 6. Leading and Trailing Aircraft Airspeed (Kts) versus Time (Sec)

Bank Angle

The evolution of the controlled bank angle of the trailing aircraft is shown in the Figure 7. The choice of the value of $\lambda_{\psi 0}(0)$ allows for clean turns thanks to a rapid command of the maximum / minimum values of the bank angle, i.e. 20 degrees.

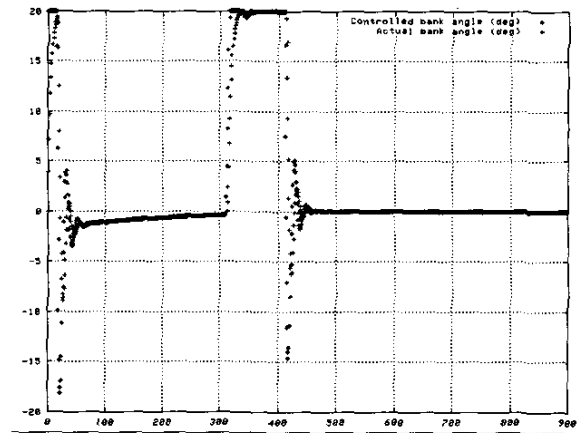


Figure 7. Controlled Bank Angle (Degrees) versus Time (Sec)

Load Factor

Finally, the evolution of the load factor for the trailing aircraft is shown in the Figure 8. It shows that the maneuver remains quite comfortable for the passengers. Indeed, maximum longitudinal acceleration for civil flights is 2 fps^2 , i.e. $1.06 \times g$ [11].

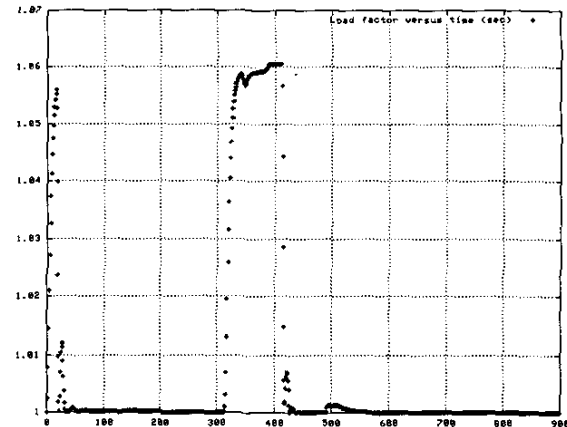


Figure 8. Load Factor versus Time (Sec)

Conclusion

In this paper, the design of a supervised controller to move towards a leading aircraft and to maintain a constant time delay behind it has been considered. A simplified case where the aircraft relative motion is limited within the horizontal plane without wind has been considered.

The proposed approach is based on backstepping. One of the key-point of such a design is the use of a skew-symmetric property of a matrix which appears in the relative motion kinematics. Furthermore, the supervisor has been designed in order to give favor to constant speed maneuver when the cross-track distance between the trailing and the leading aircraft is 'high'.

As far as available on-line information and communications are concerned, more realistic performances may be achieved by introducing wind and noisy data filtered by an observer.

This approach appears quite interesting. It requires further studies, especially on the robustness of such a controller to noisy data from the leading aircraft and to wind gusts.

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